

AIRCRAFT SPIRALLING MECHANISM - B

CROSS REFERENCE TO RELATED APPLICATIONS

This is a divisional patent application, being a division of the US patent application numbered 09/886,639.

STATEMENT REGARDING FEDERALLY SPONSORED RESEARCH OR DEVELOPMENT

5 Not applicable.

REFERENCE TO SEQUENCE LISTING

Not applicable.

BACKGROUND OF THE INVENTION

FIELD OF THE INVENTION

10 This invention is related to the field of aviation dealing with missiles and military attack airplanes.

The aim of this invention is to provide an aircraft that has higher chance of surviving attacks from anti-aircraft weapons when flying over enemy territory than aircraft
15 currently in use. The aircraft according to this invention is fitted with a mechanism that enables the aircraft to travel in a continuous spiralling motion while flying over enemy territory, without the need for a pilot to make continues control adjustments. The mechanism is such
20 that once activated, the spiralling motion is automatic.

The mechanism can also be dis-engaged by a pilot when so desired. The spiralling motion is achieved during flight without having to roll the aircraft.

While a pilot flying a conventional aircraft such as a jet fighter could make the conventional aircraft fly in a spiralling motion, this could only be achieved if the pilot kept making continuous control changes with his own arm. This could become quite tiresome and strenuous after a while and would require continued concentration, if the spiralling was achieved without rolling the aircraft. Rolling the aircraft, as in the form of a barrel roll, may seem like an easy alternative, but continuous rolling would make a pilot dizzy after a while, leading to loss of control, and if close to the ground, a potential for a crash. A continuous rolling motion would also make it hard for the pilot to observe enemy territory, navigate and make target selection.

That is, using a sustained rolling motion in order to achieve a prolonged spiralling motion would not be practical.

The aircraft in this invention would allow the pilot to operate conventional controls in a conventional manner, as when flying in a smooth manner, while the aircraft continued to travel in a spiralling motion. The advantage of this is that the pilot would be able to continue to observe enemy territory and would be free to concentrate on targeting enemy sites while the aircraft flew in an evasive manner.

DESCRIPTION OF THE RELATED ART

US Patent 5,322,243 in the name of Stoy shows a missile with variable pitch fins on a rotatable tube that are moved by independent actuators, and a computer to control 5 the operation of the actuators. While the intention of Stoy wasn't to provide a missile that could travel in a continuous spiralling motion, such a motion could be achieved by the missile shown in Stoy's patent with appropriate programming of the controlling computer. The 10 current invention provides a mechanical means for inducing a spiralling motion in an aircraft that does not need a computer to control the position of the fins on a rotatable tube to induce a spiralling motion in the aircraft.

BRIEF SUMMARY OF THE INVENTION

In this invention the spiralling motion of a fast flying aircraft is achieved by using moveable fins on a rotatable tube, with the tube encircling a part 5 of the main body aircraft and with the tube able to rotate around the encircled part of the aircraft.

The fins are attached to the rotatable tube so that they can be rotated in a pivoting manner relative to the rotatable tube. That is, if the rotatable tube was 10 kept in a fixed position on the airplane so as not to rotate, the fin movement would resemble the movement of canards on aircraft such as the Eurofighter and the recent version of the Sukhoi Su-37. The fins would turn in a pitch altering motion in the same direction. With 15 the fins horizontal, the aircraft would be allowed to fly smoothly. When the fins are rotated from the horizontal position, they would act to push the aircraft in a similar manner to the way that canards would (if positioned on the forward part of the aircraft).

For the aircraft to enter a spiralling motion, the fins would need to revolve around the body of the aircraft so that the aircraft is pushed in changing directions. In the invention this is achieved by using the rotatable tube

5 that allows the fins to revolve around the main body of the aircraft - using the rotatable tube as a means of travelling around a part of the main body of the aircraft. Fins that are of unequal size with respect to one another are used to cause the rotatable tube to rotate. Having

10 fins that are of unequal size would cause an aerodynamic imbalance when the fins are moved from the horizontal position. With one fin pushing harder than the other, rotation of rotatable tube would result. The rotation of the rotatable tube would be automatic and continuous while

15 the imbalance between the fins was maintained. Placing the fins back in a horizontal position would remove the imbalance, allowing the rotatable tube to come to rest. Friction between the aircraft and the rotatable tube or a braking mechanism such as a hydraulically activated

20 brake pad being pushed against the rotatable tube could help to stop the rotatable tube from rotating.

Another way of causing the rotatable tube to rotate is to increase the pitch of one fin more than that of the other. Increasing the pitch of one fin relative to the other would cause an aerodynamic imbalance on the rotatable 5 tube, thereby forcing it to rotate. Allowing the fins to return to a horizontal position would remove the aerodynamic imbalance, allowing the rotatable tube to come to rest.

Although the aircraft could be in the form of a jet propelled airplane, it could be in the form of any one of 10 a range of aircraft such as guided missiles and unguided missiles. It could also be in the form of unpropelled aircraft such as gliders or winged bombs that are designed to glide to a target.

BRIEF DESCRIPTION OF THE SEVERAL VIEWS OF THE DRAWINGS

15 Embodiments of the invention will now be described by way of example with reference to the accompanying drawings, of which:

FIG. 1 shows the left side view of an aircraft in the form of a jet airplane comprising a spiral inducing assembly.

FIG. 2 shows an enlarged view of the spiral inducing 20 assembly of FIG. 1.

FIG. 3 shows the left side of the spiral inducing assembly of FIG. 1 after the spiral inducing assembly has been activated to cause a spiralling motion to occur.

FIG. 4 shows the right side of the spiral inducing assembly 5 of FIG. 1.

FIG. 5 shows the airplane of FIG. 1 as viewed from underneath the airplane.

FIG. 6 shows the spiral inducing assembly of FIG. 4 in an activated state, and after the rotatable tube has been 10 rotated.

FIG. 7 shows an aircraft according to this invention in the form of a missile.

FIG. 8 shows a cross-sectional view of the spiral inducing assembly of FIG. 1 as viewed from the front of the airplane.

15 FIG. 9 shows a cross-sectional view of the spiral inducing assembly as viewed from behind the spiral inducing assembly.

FIG. 10 shows the left side of the front of the fuselage of the airplane of FIG. 1.

DETAILED DESCRIPTION OF THE DRAWINGS

FIG. 1 shows one form of the aircraft 1 as a jet propelled airplane 1, fitted with a spiral inducing assembly 2.

Referring to FIG. 1, a rotatable tube 3 forming part of 5 the spiral inducing assembly 2 can be seen encircling part of the fuselage 4 of the airplane 1. The fuselage has a fore end and aft end. Referring to this tube 3 as the primary tube 3, the primary tube 3 is able to rotate around the part of the fuselage encircled by the primary tube. The 10 primary tube is shown as being narrower in the front than at the rear. Also shown is another tube 5 that is fitted to the airplane such that it encircles part of the fuselage 4 of the airplane. Referring to this tube 5 as the activation tube 5, the activation tube 15 is fitted so that it can be moved in a forward direction relative to the part of the fuselage 4 encircled by the activation tube and then back to its original position on the fuselage. FIG. 1 also shows the edge of one horizontal fin 6 that is connected to 20 the outside of the primary tube 3. The fin 6 is connected to the outside of primary tube 3 such that it can rotate in a pivoting manner as shown in FIG. 3.

FIG. 2 shows an enlarged illustration of the left side of the spiral inducing assembly 2. The fin 6 in FIG. 2 is connected to the outside of the primary tube 3 by a connecting joint which is 5 in the form of a connecting rod 7. Extended from the connecting rod 7 in FIG. 2 is a protruding section 8 which is used to rotate the connecting rod 7. Rotation of the connecting rod 7 causes the fin 6 to rotate in a pivoting manner around the 10 connecting rod 7 (in the manner shown in FIG. 3). Linked to the protruding section 8 in FIG. 2 is a stem 9. Referring to this stem 9 as an activation stem 9, the activation stem 9 is used as a means for pushing the protruding section 8 such that when the 15 protruding section 8 is pushed, the protruding section 8 forces the connecting rod 7 to rotate around the longitudinal axis of the connecting rod 7. The activation stem 9 is linked to the protruding section 8 by a rivet 10. The activation stem 9 is 20 shown as being fitted on the inside of the primary tube 3 and is supported inside the primary tube 3 by a retaining bracket 11. The retaining bracket 11 is rigidly joined to the inside of the primary tube but is channelled to allow the activation stem 9 to move longitudinally 25 between the retaining bracket 11 and the primary tube 3. The activation stem 9 is allowed to protrude

rearward from the primary tube so that it can be reached by the activation tube 5 when the activation tube 5 is moved forward on the fuselage 4. The activation tube 5 is forced to move forward by an 5 activation mechanism 12 consisting of hydraulic actuators 13 and 14.

FIG. 3 shows that as the activation tube 5 is forced to move forward on the fuselage 4 when the hydraulic actuators 13 and 14 extend, it eventually makes 10 contact with the activation stem 9. As the activation tube 5 is forced to move further forward, it pushes the activation stem 9 forward on primary tube. As the activation stem 9 is pushed forward, the activation stem pushes against the protruding section 8 and moves 15 the protruding section 8, thereby rotating the fin 6 around the connecting rod 7 in a pivoting manner.

In FIG. 3 a rivet 10 is shown connecting the activation stem 9 to the protruding section 8, which allows movement between the activation stem 9 and the 20 protruding section 8. The retaining bracket 11 keeps the activation stem from moving laterally around the primary tube. The retaining bracket 11 however does allow longitudinal sliding movement of the activation stem 9 so that it can be pushed and moved by the 25 activation tube 5.

FIG. 4 shows the right side of the spiral inducing assembly 2 of FIG. 1. Shown is another fin 17, another connecting joint in the form of a connecting rod 18 that connects the fin 17 to the outside of the primary tube 3. Another protruding section 19 is used to rotate the connecting rod 18, and the activation stem 20 is used to push the protruding section 19, with the activation stem 20 linked to the protruding section 19 by a rivet 21. Also visible in FIG. 4 is the activation tube 5. The connecting rod 18 allows the fin 17 to rotate in a pivoting manner. Another retaining bracket 22 is shown supporting the respective activation stem 20.

FIG. 4 shows the hydraulic actuators 15 and 16 located on the right side of the spiral inducing assembly 2 which also form part of the activation mechanism 12 by which the activation tube 5 is forced to move. When the hydraulic actuators 13 14 15 and 16 are forced to extend as hydraulic pressure is applied to them, they force the activation tube 5 to move forward as shown in FIG. 3.

Thus, it can be seen from FIG.s 1, 2, 3 and 4 that the activation tube 5, the activation stems 9 and 20, retaining brackets 11 and 22, protruding sections 8 and 19, rivets 10 and 21 used to connect the 5 activation stems 9 and 20 to respective protruding sections 8 and 19, the connecting joints 7 and 18 in the form of connecting rods 7 and 18, and the activation mechanism 12 used to move the activation tube 5 consisting of the hydraulic actuators 13, 14, 15 and 16, collectively form a 10 fin rotating mechanism, by which fin rotating mechanism the fins can be rotated symmetrically in the same direction, so that the force exerted on the primary tube by one fin can be overcome by the force exerted on the primary tube by another fin.

FIG. 5 shows the airplane 1 of FIG. 1 from underneath. It shows that one fin 6 is larger than the other fin 17. When these fins 6 and 17 are rotated in a pivoting manner and in the same direction to the same extent, an aerodynamic 5 imbalance between the fins 6 and 17 arises during flight of the airplane because of size difference between the fins 6 and 17. The larger fin 6 will exert a greater magnitude of force on the primary tube 3 during flight of the airplane 1 than the smaller fin 17. As a result, the aerodynamic 10 imbalance between the fins 6 and 17 would cause the primary tube 3 to rotate. But both fins 16 and 17 would also be pushing the airplane laterally, in a similar manner to canards. Thus, because the primary tube 3 is forced to 15 rotate, the lateral force exerted on the airplane by the fins 6 and 17 keeps changing, thus forcing the airplane to keep changing its direction and hence entering a spiralling motion.

FIG. 6 shows the spiral inducing assembly of FIG. 4 with the fins 6 and 17 of FIG. 4, and with the primary 20 tube 3 in a state of rotation. It can be seen comparing FIG. 5 with FIG. 6 how the lateral forces on the airplane would be constantly changing, enabling the spiral inducing assembly 2, to force the airplane 1 to travel in a continuous spiralling motion.

Looking at the fins 6 and 17 shown in FIG. 6 it can be seen that the rear section of each fin behind the respective connecting rods 7 and 18 is greater than the section of each fin in front the respective connecting 5 rods 7 and 18. This is deliberate. This is used to allow the fins to adopt a horizontal position when hydraulic pressure is released from the hydraulic actuators 13, 14 (and 15 and 16 of FIG. 4) allowing the activation tube 5 to retreat away from the 10 primary tube 3. Aerodynamic forces are in effect used to allow the fins to return to a resting horizontal position, allowing the airplane to re-commence a smooth 15 FIG. 7 shows an aircraft according to this invention in the form of a missile with a spiral inducing assembly 2 15 of FIG. 1.

FIG. 8 shows a cross-sectional view of the spiral inducing assembly of FIG. 1 as viewed from the front of the airplane. Shown here is the primary tube 3, the fins 6 and 17, (with fin 6 being larger than fin 17), the fuselage 4 of the 20 airplane, the activation stems 9 and 20, linked by rivets to the protruding sections 8 and 19 respectively, the connecting rods 7 and 18 penetrating the primary tube 3, and with the protruding sections 8 and 19 screwed in the connecting rods 7 and 18 respectively. FIG. 8 shows a way of supporting the 25 primary tube 3. Shown is a tube of smaller diameter 23 than the primary tube 3. This smaller tube 23 is a supporting

tube in that it is used to support the primary tube 3. It has a smaller diameter than the primary tube 3 to provide a gap 24 between the primary tube 3 and the supporting tube 23. The gap 24 is used to allow 5 freedom of movement to the protruding sections 8 and 19, and the activation stems 9 and 20 shown positioned inside the primary tube 3. Bolts 25, 26, 27 and 28 are used to join the primary tube 3 to the supporting tube 23. The supporting tube 23 is able to rotate around 10 the encircled part of the fuselage 4.

FIG. 9 shows a cross-sectional view of the spiral inducing assembly as viewed from behind the spiral inducing assembly. Shown in FIG. 9 are the rear ends of the activation stems 9 and 20, and the retaining brackets 11 and 22 that support 15 the activation stems 9 and 20, and prevent uncontrolled lateral movement of the activation stems 9 and 20.

FIG. 10 shows a side cutting of the part of the fuselage 4 encircled by the primary tube 3 of FIG. 1. The encircled part of the fuselage 4 can be seen to be narrower 20 than the rest of the fuselage 4. Thrust bearings 29 and 30 are positioned on the narrowed section of fuselage 4. The thrust bearings are used to prevent the primary tube moving longitudinally relative to the fuselage 4.